

Yaw angle effect on the aerodynamic performance of hatchback vehicle fitted with wing spoiler

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Abstract

Research on spoiler available to date was mainly done to optimize the performance of spoiler in non-zero yaw condition. However, the effect of spoiler is most needed during cornering to ensure the stability of the vehicle. Therefore, this study aims to inspect the effect of yaw angles change on the aerodynamic performance of the NACA 0018 wing spoiler and the subsequent influence on the flow characteristics of the hatchback vehicle. Computational Fluid Dynamics (CFD) has been applied to model the flow. Comparison between numerically obtained results and experimental data was done to validate the CFD method. The findings show that both the drag coefficient, C_d , and lift coefficient, C_l have increased with increasing yaw angle. However, the spoiler has performed in favor of reducing the C_d and C_l even with increasing yaw angle. The averaged proportion contributions from the spoiler to the overall C_d and C_l are 2.7% and 4.1%, respectively. The other body parts that have contributed to the C_d and C_l reductions were the base and slant, and the roof.

Keywords: Yaw angle, spoiler, aerodynamics

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INTRODUCTION

During cornering, a vehicle needs sufficient frictional forces in between its tires and the road surface to develop sufficient centripetal force to pass the curve without slip. The frictional forces are directly proportional to the downward forces exerted on the vehicle. In principle, the downward forces depend on the weight of the vehicle and the downward component of the aerodynamic force. Although, it may seem intuitive to increase the weight of the vehicle to enhance the frictional force, but this will not improve the performance of the vehicle during cornering due to the fact that the additional weight will produce additional inertia for which the vehicle will need to overcome. Besides, a heavier vehicle will result in higher fuel consumption rate, which is a disadvantage for both economic and environmental considerations. Therefore, the only viable solution would be to rely on aerodynamic effects.

The aerodynamic effects can be obtained by properly installing the aerodynamic devices on the vehicle. One of the common aerodynamic devices is known as spoiler. In general, there are two types of spoilers, front and rear spoilers. A front spoiler is attached to the bumper while a rear spoiler is usually attached to the trailing edge of the roof for hatchback vehicles. The function of a rear spoiler is, as the name implies, to 'spoil' the smooth airflow passing over a vehicle in order to alter its aerodynamic characteristics, such as to reduce drag and increase downforce.

Spoilers of various kinds had been widely researched (S. Cheng & S. Mansor, 2017), S. Y. Cheng and S. Mansor (2017), Hu and Wong (2011), Mitra (2010), and Kieffer, Moujaes, and Armbya (2006). To date, numerous studies concerning the optimization of the performance of wing spoilers have been reported in the literature. For example, Tsai, Fu, Tai, Huang, and Leong (2009) have studied the effect of various rear spoiler designs which were of aerofoil profiles on the aerodynamic

characteristics of a simplified car model. It shows a reduction in lift. Besides, the study of Kim, Chen, and Shulze (2006) shows that by using a spoiler on a minivan, the lift force can be reduced by 100% when the minivan is driving at a speed of 108km/h. Moreover, the studies conducted by Daryakenari, Abdullah, Zulkifli, Sundararajan, and Sood (2013) and Kodali and Bezavada (2012) have also shown the similar lift reduction tendency produced by the spoilers. Although the studies on optimizing the performance of wing-type spoilers are not uncommon, but most of them are limited to zero degree yaw condition (i.e. simulating a vehicle driving in a straight path). However, during cornering, the effect of spoilers in enhancing the downforce is most needed to ensure drive stability and safety through the curve. Therefore, to address the issue, the main objective of the present study is to investigate the aerodynamic performance of the wing spoiler in yawing conditions.

The paper is organized as follows: the chosen hatchback vehicle model is first introduced together with the features of the spoiler. In addition, the convention of yaw condition is briefly explained. Section II documents the CFD settings and validation of the numerical method. In Section III, the results obtained are being reported and discussed. The paper ends with the conclusion in Section IV.

METHODOLOGY

Hatchback model and spoiler configuration

In this study, the Ahmed body, a reference road vehicle model introduced by Ahmed, Ramm, and Falin (1984) is adopted for simulating the hatchback-type vehicle. The 35° slant angle of the upper rear section of the model is chosen which is typical for most hatchback vehicles.

The Ahmed body fitted with a rear-roof NACA 0018 wing spoiler with 5° angle of attack is shown in Fig. 1. The chord length of the wing

is 69 mm. The chord length was decided based on the measurements done on the wing spoiler of real cars and scaled down based on the dimensions of the Ahmed model.

Meanwhile, in order to avoid highly skewed cells during meshing, the sharp end of the wing has been filleted (2 mm radius). For the same reason, all the sharp edges of the two endplates supporting the wing spoiler were filleted (2 mm radius). For the details of the Ahmed model dimensions, the reader is referred to Ahmed *et al.* (1984).

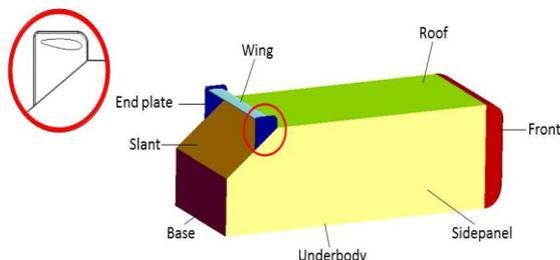


Fig. 1 Ahmed model fitted with a NACA 0018 wing spoiler.

Fig. 2 shows the convention of yaw angles adopted in this study. In the present study, the yaw angles investigated were from 0° to 12°, at 4° increment. According to vehicles in real life, the yaw angle will not go beyond of 12° during cornering. Thus we limit the investigation to 12° maximum yaw angle.

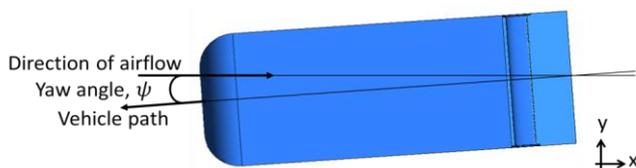


Fig. 2 Convention of yaw angle.

Computational fluid dynamics (CFD) settings

A numerical simulation method is used in this study to investigate the influence of varying the yaw angle on the aerodynamics of a hatchback model mounted with a NACA 0018 wing spoiler. The commercial CFD software was utilized in the present study. It is a pressure-based solver based on the finite-volume method known as ANSYS Fluent. In fact, ANSYS’s CFD solver had been widely used and validated in various applications. For instances, in fields like sports (Jahi, Zawawi, & Rahman, 2015), medical(Missel, Horner, & Muralikrishnan, 2010), combustion (Torabmostaedi, Zhang, Foot, Dembele, & Fernandez, 2013), heat transfer (Alawi, Sidik, & Tey, 2015; Munir, Mohd Azmi, Razali, & Tokit, 2012), urban aerodynamics (e.g. Cheng, 2007), and automotive (Shafie, Kamar, & Kamsah, 2015a, 2015b). In the CFD modelling, the time-averaged, incompressible, continuity and momentum equations are solved as follow:

$$\frac{\partial u_i}{\partial x_i} = 0 \tag{1}$$

$$\rho u_j \frac{\partial u_i}{\partial x_i} = \frac{\partial}{\partial x_i} [-p\delta_{ij} + 2\mu S_{ij} - \rho u'_i u'_j] \tag{2}$$

- where, ρ = fluid density
- u = time-averaged velocity
- p = time-averaged static pressure
- μ = viscosity of fluid
- S_{ij} = mean strain tensor rate
- $\rho u'_i u'_j$ = Reynolds stresses representing the effects of turbulence.

The Reynolds stresses term resulted from performing Reynolds averaging was modelled by the k-epsilon realizable model, which is a two-equation turbulence model. Moreover, the enhanced wall treatment (EWT) has been employed. The EWT models the boundary layer

profile in the vicinity of the wall by the wall function when the first grid point is within the log-layer range. As for spatial discretization, second-order upwind difference scheme has been employed.

At domain’s inlet, a uniform velocity $U = 40$ m/s has been imposed. Its turbulence intensity was at 0.2%. The corresponding Reynolds number (Re) was 768,000 based on the model height.

A rectangular flow domain has been adopted. The domain has a cross-sectional area of 1450 mm x 3890 mm (height x width). Hence, there was only less than 2% of blockage ratio obtained, which is far lower compared to the typical accepted 5% blockage ratio (Hucho & Sovran, 1993). The inlet and outlet boundaries were respectively located at 1.4*l* upstream and 11.5*l* downstream of the model (*l* is the model length). Meanwhile, the outlet was set as pressure outlet boundary condition with zero gauge pressure. The surfaces of the model and the floor were defined as no-slip wall while the remaining walls were set as symmetry boundary condition.

Meshing

As shown in Fig. 3, the domain mainly comprises tetrahedral cells in combination with prismatic cells on the no-slip walls. The purpose of the prismatic cells is to improve the resolution of the boundary layer around the model and the ground. The outcome of grid independence test reveals that the mesh with cell number of around 1530368 is sufficiently refined. With the prismatic cells, the thickness of the first node from the model’s surface was at 0.5 mm, this corresponds to y^+ values ranging from 2 to 53.

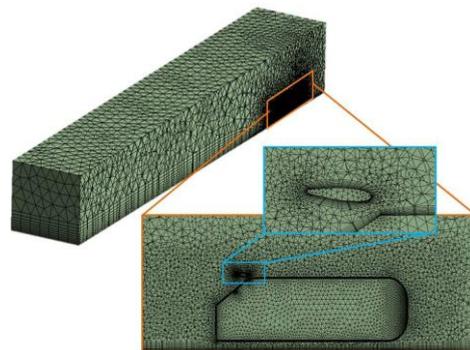


Fig. 3 Mesh density distribution around the Ahmed model with a closed-up of prismatic cells around the wing spoiler.

Validation

In order to validate the numerical method, the present study attempted to reproduce the experimental results by Bello-Millán, Mäkelä, Parras, del Pino, and Ferrera (2016). Fig. 4 compares the drag coefficient C_d of the Ahmed model at increasing yaw angle obtained by the present CFD method and the experimental work by Bello-Millán *et al.* (2016). As may be seen, the two curves are in very good agreement (with the maximum difference of 5.2% at 20° yaw angle).

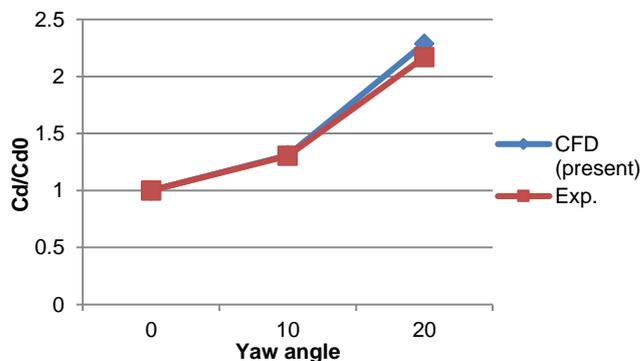


Fig 4 Effect of yaw angle on the C_d of Ahmed model obtained by the present CFD and the experimental work by (Bello-Millán *et al.* (2016)). The C_d values by each method are normalized by their respective C_d values at 0° yaw.

As mentioned earlier, this study only covers the yaw angle range up to 12°. Thus, by interpolation, the corresponding percentage difference between the CFD and experimental results at 12° yaw angle is only 0.6%. Therefore, we can deduce that the present CFD method is suitable for the investigation.

RESULTS AND FINDINGS

Effect of yaw angle on C_d and C_l

Fig. 5 shows the effect of yaw angle on the total C_d and C_l of the model. As may be seen, the two aerodynamic force coefficients have risen with increasing yaw angle (Note that the C_d is defined as the aerodynamic force component parallel to the longitudinal axis of the model). Thus, the results suggest when the vehicle is no longer travelling in a straight path, its aerodynamic performances will deteriorate.

Fig. 6 shows the effect of yaw angles on the C_d and C_l of the spoiler. A decreasing trend is observed in the two aerodynamic quantities. In particular, from 0° to 12° yaw angle, both the C_d and C_l have decreased by about 22% and 43%, respectively. The decreasing trend suggests that the spoiler will actually function better at increasing yaw angle.

However, since the values of the two force coefficients of the spoiler are relatively small compared to the other body parts of the model, thus it failed to alter the increasing trend of the overall C_d and C_l . In fact, the proportion contribution of the wing spoiler to the overall C_d was only 3.24% at 0° yaw. As the yaw angle increased to 12°, the proportion contribution of the spoiler has diminished to 2.17% due to higher increment rate of other body parts.

Meanwhile, in regard to C_l , only the underbody and wing spoiler had generated the downforce, whereas the other body parts had generated positive lift. In addition, despite the spoiler's positive influence to the overall C_l , its proportion is only about 4.1% of the overall downforce generated.

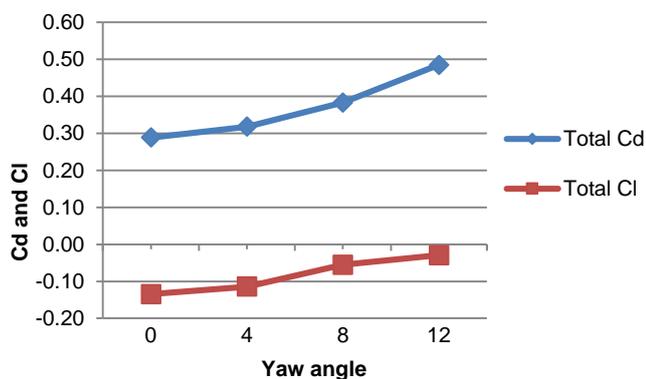


Fig. 5 Effect of yaw angles on C_d and C_l of Ahmed model with wing spoiler.

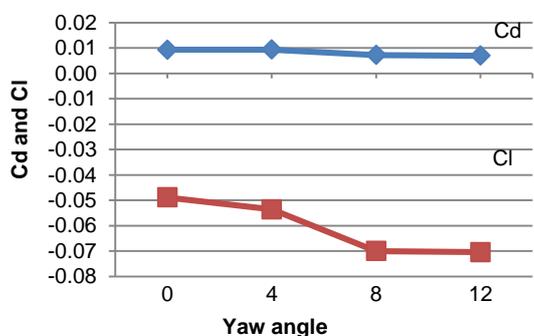


Fig. 6 Effect of yaw angles on C_d and C_l of rear wing.

Fig. 7 and 8 respectively show the C_d and C_l of each body part as a function of yaw angle. In Fig. 7, it is apparent that the main contributors to the increasing C_d were the base and slant. On the other hand, the front has an opposite, desirable influence on the C_d value.

In Fig. 8, the roof has appeared to be the main contributor to the increasing C_l tendency, whereas the underbody has exhibited an opposite influence.

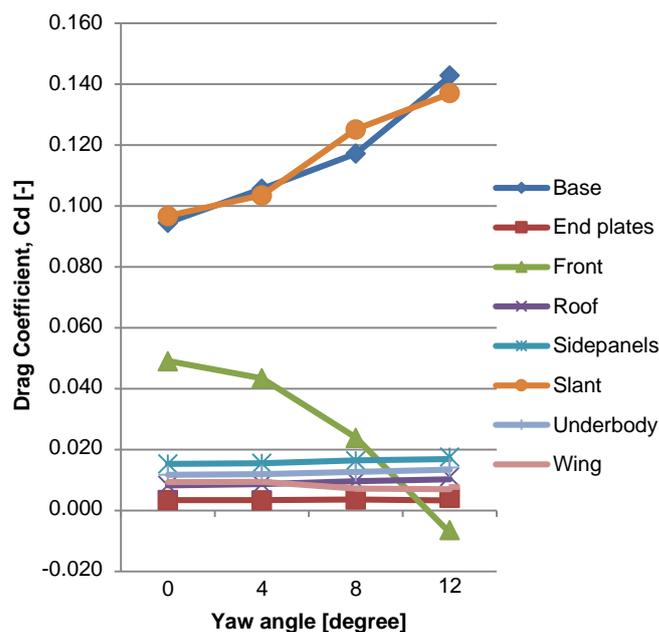


Fig. 7 Contributions of body parts to C_d with increasing yaw angles.

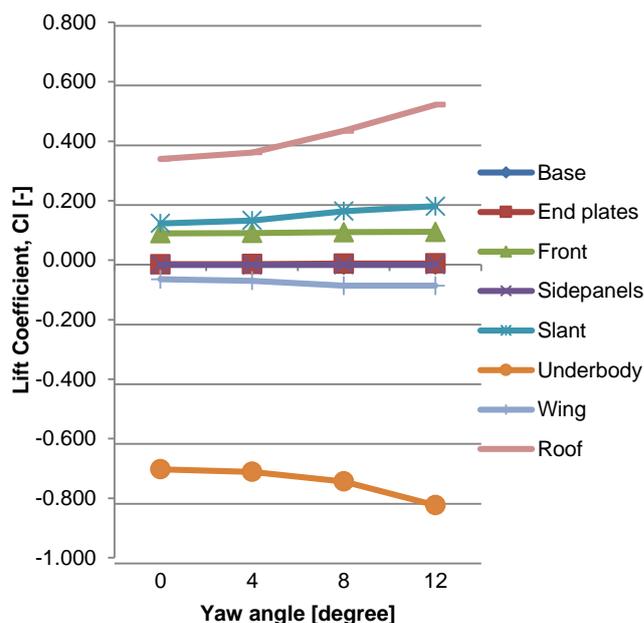


Fig. 8 Contributions of body parts to C_l with increasing yaw angles.

Effect of yaw angle on flow structures

As shown in Fig. 9, at non-zero yaw angle, four longitudinal vortices (marked A, B, C and D) were developed from the four front corners of the model. This phenomenon is considered reasonable as the front corner edges are of angular shape. Note that the vortices A and D have the counter clockwise rotation whereas the vortices B and C have the clockwise rotation.

As may be seen, the windward vortices A and B have each developed a path located on the top and bottom of the model respectively, near the windward side. On the other hand, the leeward vortices C and D have each developed a path locating away from the model surfaces. As a result of their closer distance to the model, at increasing yaw angle, the windward vortices A and B could have stronger influence on the surface pressure of the model due to the increasing vortex strength.

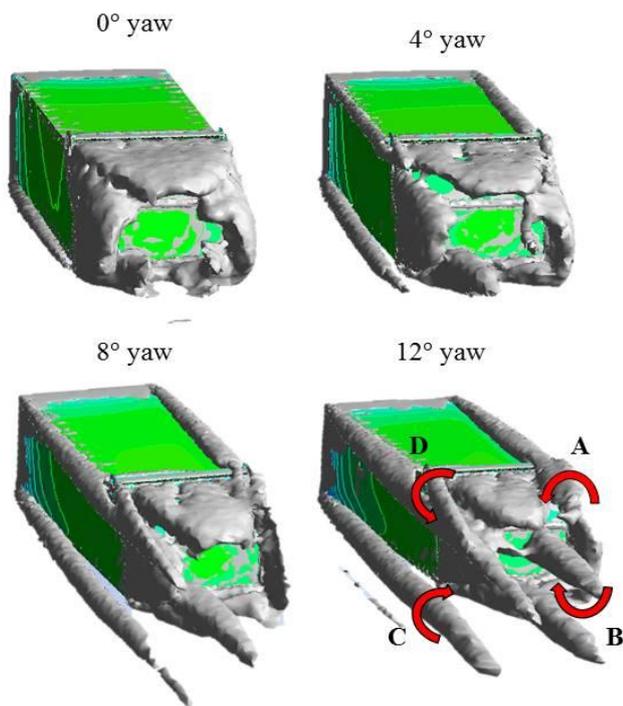


Fig. 9 Effect of yaw angle on body part contributions to C_d and C_l .

Physical mechanism

Fig. 10 compares the surface pressure distribution on the top and bottom parts of the model at different yaw angles. As illustrated, the model have shown the low surface pressure regions (regions in the red boxes) along its top and bottom side edges due to the windward vortices. On the top part of the model, a decrease in the pressure of the roof could cause an increase in the C_l . Hence, the roof of the model has shown the increasing C_l trend with increasing yaw angle. In fact, the result in section 3.1 shows that the roof is the main contributor to the increasing trend of the overall C_l with increasing yaw angles.

Meanwhile, for the bottom part of the model, a decrease in the surface pressure will cause the C_l to decrease. Thus, the underbody has exhibited the favourable influence of increasing the downforce with increasing yaw angles.

The reason for the reduction in the C_d of the front part is illustrated in Fig. 11. As may be seen, the low surface pressure region at the leeward side (i.e. left vertical edge) has become wider with increasing yaw angle. This phenomenon was caused by the flow acceleration near the leeward side corner at higher yaw angle as shown in Fig. 12.

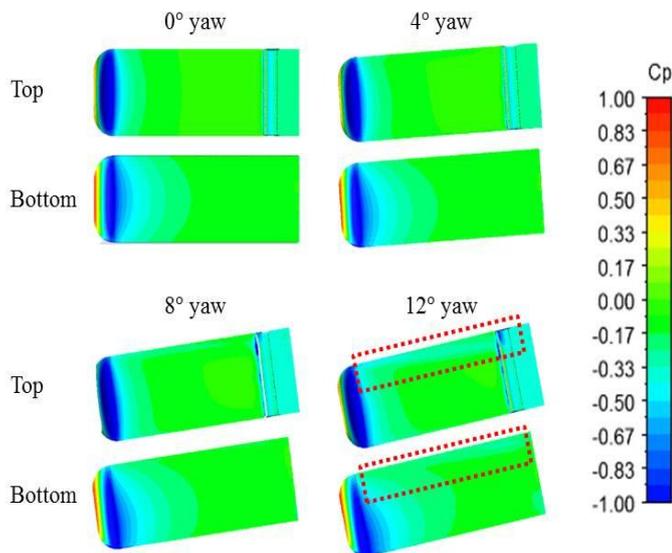


Fig. 10 Surface pressure distributions at different yaw angles (top view).

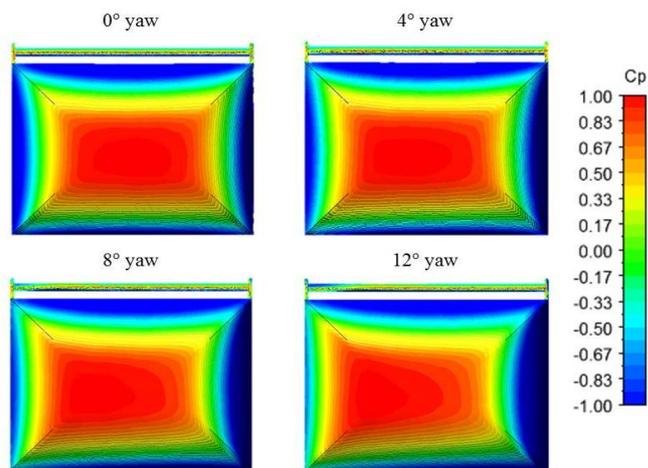


Fig. 11 Surface pressure distribution of the front part at different yaw angles (front view).

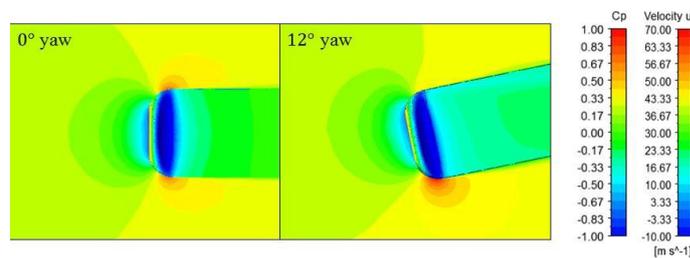


Fig. 12 Yaw angle effect on the velocity distribution near the front part of the model (top view; visualization plane at the mid height of the model).

On the other hand, the surface pressure drop of the slant and base at higher yaw angles (see Fig. 10 and Fig. 13) is the reason why the two body parts had caused the increase in the model’s overall C_d at higher yaw angles. Their pressure drop is associated with the increase in the size of the separation bubbles at the back of the model at higher yaw angle (see Fig. 13).

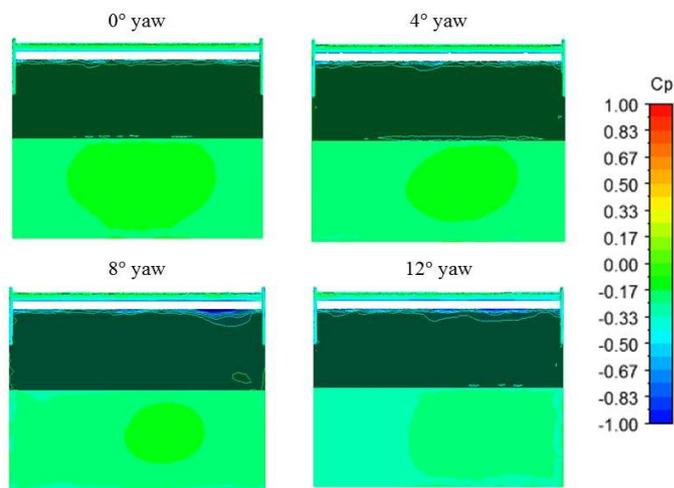


Fig. 13 Surface pressure distributions at different yaw angles (back view).

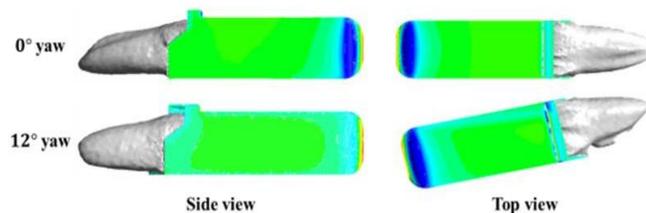


Fig. 14 Separation bubbles formed at different yaw angles.

CONCLUSION

In the present study, a RANS-based CFD modelling has been used to study the influence of varying the yaw angle on the aerodynamic characteristics of an idealized hatchback reference body fitted with a NACA 0018 wing spoiler. The results obtained indicate that both the C_d and C_l of the simplified vehicle model have increased with increasing yaw angle. This tendency is deemed unfavourable to fuel economy and drive safety. Although the C_d and C_l of the wing spoiler have shown decreasing trends, however, their contribution to the overall C_d and C_l are small, only with the averaged proportion contributions of 2.7% and 4.1% respectively. Hence, the main contributors to the increase of C_d and C_l had overshadowed the effects brought about by the wing spoiler. The main body parts that have contributed to the increase in C_d at higher yaw angle are the base and slant, while the rise in C_l is caused by the roof.

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